## Reply by Author to C. M. Hung and C. Fan

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**H**UNG and Fan's<sup>1</sup> improvement upon our analysis is a welcome one and significant whenever the detailed disturbance field away from the wall is of interest; indeed, it further strengthens our own earlier assertion<sup>2</sup> as to the importance of lateral pressure gradient effects in the study of small disturbances within highly nonuniform boundary-layer flows. We would like to re-emphasize, however, that our solution is exact near the wall (it is readily shown to be identical with Hung and Fan's theory as  $y \rightarrow y_f$ ), gives acceptable agreement with experiment as Ref. 1 shows, and has proven a valuable ingredient in the theoretical analysis of other important problems such as heat and mass transfer disturbances,<sup>3</sup> ablation surface cross-hatching phenomena,<sup>4</sup> and liquid film stability in high-speed gas streams.<sup>5</sup>

Secondly, it is pointed out that the apparent disagreement in Fig. 4 of Ref. 1 regarding the phase angle variation is due entirely to a different choice of the reference angle: in our work, the local pressure phase was measured relative to the corresponding *local* maximum slope point pertaining to a wavy wall at this y, whereas Hung and Fan's pressure phasing is respect to the *fixed* location  $y = y_f$ . When this difference is accounted for, the phase angles given by the two analyses are the same; the only difference between them is the amplitude results for  $y \gg y_f$ .

Third, some remarks on the suitable choice of an analytical model of the Mach number (velocity) profile are deemed appropriate here. Owing to the presence of the  $\partial M/\partial y$  term in the Lighthill pressure perturbation equation [their Eq. (1), our Eq. (2)], the velocity profile must be carefully chosen to yield a slope variation which is both continuous and nonsingular across the boundary layer while vanishing at its edge. The one-seventh power law profile used in the turbulent case by Hung and

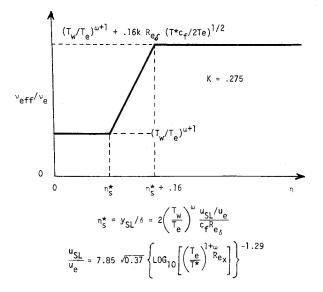


Fig. 1 Analytical model of turbulent boundary-layer profile.

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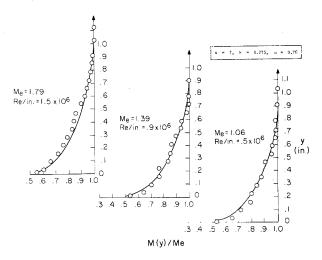


Fig. 2 Comparison of profile model with experiments of Inger and Williams.<sup>2</sup>

Fan does not satisfy the latter requirement and furthermore becomes inaccurate in precisely the turbulent law of the wall region where the effective wall conditions must be applied at  $y = y_f$ . Use of such a profile can give inaccurate solutions of the Lighthill equation as we found to be the case in our own earlier studies; consequently, we used instead a more realistic profile model<sup>6</sup> that obeys the proper law of the wall behavior for small y while giving a vanishing  $\partial M/\partial y$  at  $y = \delta$ . This model is described by the following equations:

$$\frac{d(u/u_e)}{d\eta} \simeq \frac{c_f}{2} Re_{\delta} \frac{T}{T_e} \left( \frac{\tau/\tau_w}{v_{\text{eff}}/v_e} \right) \tag{1}$$

where

$$c_f \simeq 3.70 \frac{T_e}{T^*} \left\{ \log_{10} \left[ \left( \frac{T_e}{T^*} \right)^{1+\omega} Re_x \right] \right\}^{-2.58}$$
 (2)

$$\frac{Re_{\delta}}{Re_{x}} = \frac{(n+2)(n+3)}{2n} c_{f} \left[ 1 + \frac{n(\gamma-1)M_{e}^{2}}{2(n+1)(n+2)} + \frac{T_{w} - T_{e}}{(n+1)T_{e}} \right]$$
(3)

$$\frac{T^*}{T_e} \simeq 1 + 0.044 \, M_e^2 + 0.50 \left( \frac{T_w}{T_e} - 1 \right) \tag{4}$$

with T given by the Crocco energy equation integral,  $\eta \equiv y/\delta$ ,  $\tau/\tau_w \simeq 1-3\eta^2+2\eta^3$ , and  $v_{\rm eff}$  is the piecewise-continuous effective eddy viscosity function shown here in Fig. 1. As illustrated in Fig. 2 of the present Note this model gave good agreement with experimental turbulent Mach number profiles. The lack of agreement of Hung and Fan's Fig. 3 amplitude results with ours near the wall (where in fact they should be identical) is probably attributable to the inaccuracies associated with their profile choice.

## References

<sup>1</sup> Hung, C. M. and C. Fan, "Compressible Boundary Layer Flow Over a Wavy Wall," *AIAA Journal*, Vol. 13, No. 2, Feb. 1975.

<sup>2</sup> Inger, G. R. and Williams, E. P., "Subsonic and Supersonic Boundary Layer Flow Past a Wavy Wall," *AIAA Journal*, Vol. 10, May 1972, pp. 636–642.

<sup>3</sup> Inger, G. R., "Compressible Boundary Layer Flow Past a Swept Wavy Wall with Heat Transfer and Ablation," *Astronautica Acta*, Vol. 16, 1971, pp. 325–338.

<sup>4</sup> Grabow, R. M. and White, C. O., "A Surface Flow Approach for Predicting Cross-Hatched Patterns," *AIAA Journal*, Vol. 11, June 1973, pp. 841–847.

<sup>5</sup> Bordner, G. L. and A. M. Nayfeh, "Nonlinear Stability of a Liquid Film Adjacent to a Viscous Supersonic Stream," Rept. E-74-11, April 1974, Virginia Polytechnic Institute and State University, Blacksburg, Va.

<sup>6</sup> Williams, E. P. and Inger, G. R., "Final Report: Investigations of Ablation Surface Cross-Hatching," SAMSO TR-70-246, June 1970, McDonnell-Douglas Astronautics Co.—West, Los Angeles, Calif.